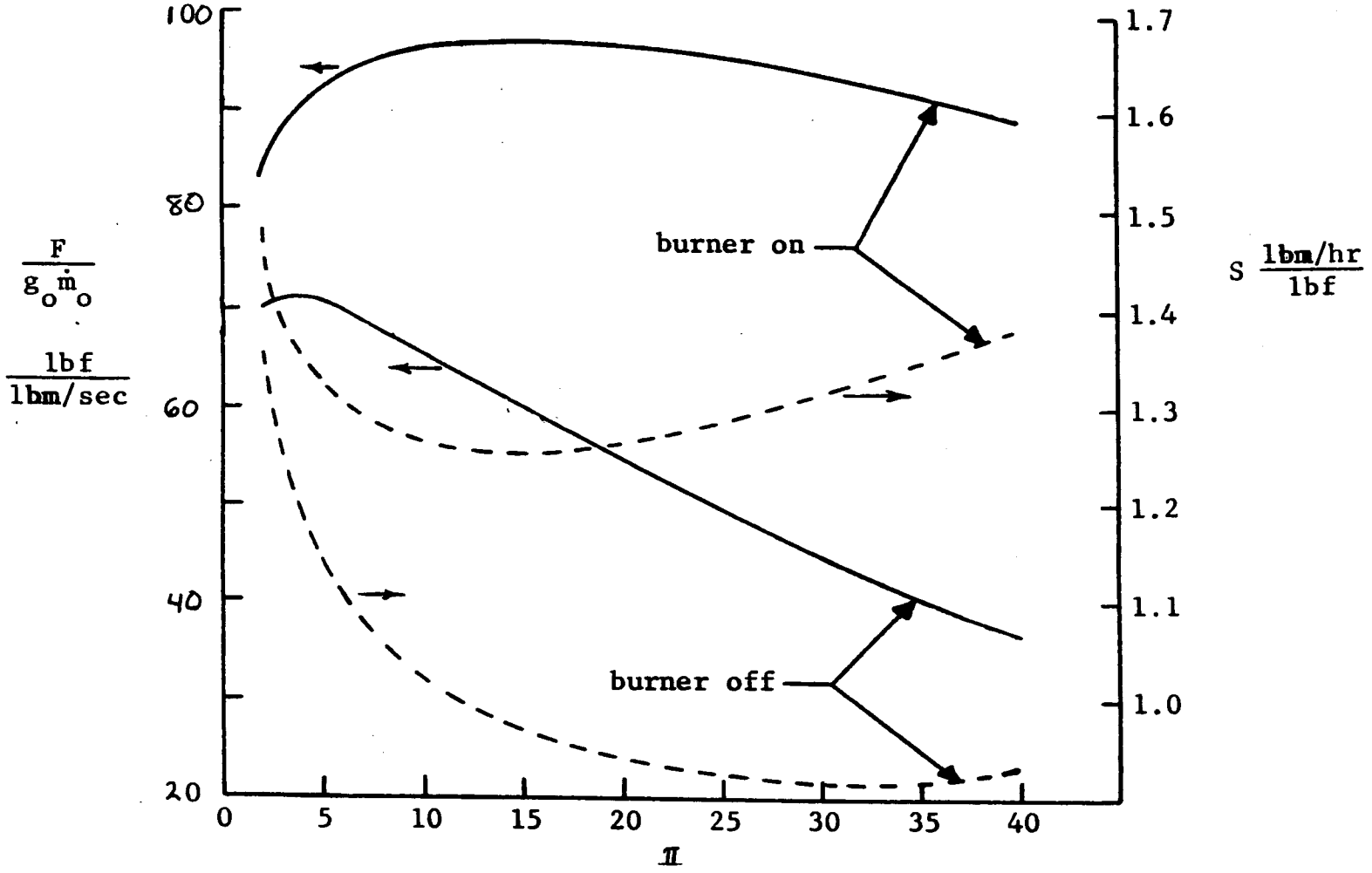

Specific thrust and specific fuel consumption for turbojet with afterburner.

Non-ideal engine with parameters: T_a = 420 R (233 K), Q_R = 19,500 Btu/lbm (45,354 kJ/kg), γ_c = 1.4, γ_t = 1.35, c_{pc} = 0.238 Btu/lbm·R (0.996 kJ/kg·K), c_{pt} = 0.262 Btu/lbm·R (1.097kJ/kg·K).

Pressure losses: π_d = 0.9425, π_b = 0.98, $\pi_n\pi_{AB}$ = 0.96 (ab on), $\pi_n\pi_{AB}$ = 0.98 (ab off).

Compressor/turbine polytropic efficiencies: $e_c = 0.92$, $e_t = 0.91$.

Combustion/mechanical efficiencies: $\eta_b = 0.97$, $\eta_m = 0.99$.